

Screech tones from rectangular jets with spanwise oblique shock-cell structures

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Understanding screech is especially important for the design of advanced aircraft because screech can cause sonic fatigue failure of aircraft structures. Although the connection between shock-cell spacing and screech frequency is well understood, the relation between non-uniformities in the shock-cell structures and the resulting amplitude, mode, and steadiness of screech have remained unexplored. This paper addresses the above issues by intentionally producing spanwise (larger nozzle dimension) variations in the shock-cell structures and studying the resulting spanwise screech mode. The spanwise-oblique shock-cell structures were produced using imperfectly expanded convergent–divergent rectangular nozzles (aspect ratio = 5) with non-uniform exit geometries. Three geometries were studied: (a) a nozzle with a spanwise uniform edge, (b) a nozzle with a spanwise oblique (single-bevelled) edge, and (c) a nozzle that had two spanwise oblique (double-bevelled) cuts to form an arrowhead-shaped nozzle. For all nozzles considered, the screech mode was antisymmetric in the transverse (smaller nozzle dimension) direction allowing focus on changes in the spanwise direction. Three types of spanwise modes were observed: symmetric (I), antisymmetric (II), and oblique (III). The following significant results emerged: (i) for all cases the screech mode corresponds with the spanwise shock-cell structure, (ii) when multiple screech modes are present, the technique presented here makes it possible to distinguish between coexisting and mutually exclusive modes, (iii) the strength of shocks 3 and 4 influences the screech source amplitude and determines whether screech is unsteady. The results presented here offer hope for a better understanding of screech and for tailoring shock-containing jets to minimize fatigue failure of aircraft components.

1. Introduction

1.1. *Motivation for the present work*

Rectangular supersonic jets with oblique or variable exits have applications in the propulsive systems of modern aircraft. The shock-cell structures and screech tone characteristics of jets from such unusual nozzles are unknown even though they significantly affect the design of nozzles and aircraft structures. Hay & Rose (1970) have documented that screech occurring in full-scale flight can damage the tailplane structure of an aircraft. They confirmed that the damage to a British Aircraft Corporation's VC 10 aircraft was indeed due to screech and not due to aerodynamic buffeting during reverse thrust. Of most concern was their finding that when the screech frequency matched a structural resonance frequency, the life-span of that structure was significantly reduced. Seiner, Manning & Ponton (1988) studied twin-jet screech resonance as it pertains to the high-level dynamic loads in the inter nozzle

region of aircraft such as the F-15 and B-1B of the United States Air Force. They also found that the dynamic loads associated with the screech resonance can reach levels capable of causing structural damage. Thus, to minimize potential structural damage, it is necessary that we understand the screech characteristics of rectangular jets from non-uniform geometries.

Interest in non-uniform jet exit geometries dates back to the early developmental stages of the Concorde (Westley & Lilley 1952; Ffowcs Williams, Simson & Virchis 1975; Smith 1989; Lilley 1991). During noise reduction attempts with the Concorde, the Olympus engines were fitted with innovative variable-geometry intake and exhaust nozzle assemblies. The variable exhaust nozzle could either close completely to reverse engine thrust or be only slightly closed to form a notch, thus squeezing the jet and altering the exit geometry. The exhaust flow from the nozzle became squeezed in the horizontal direction. The resultant flow asymmetry had little effect on the thrust but lowered the noise radiated to the side of the aircraft by 2–3 dB. This led to considerable research work on notched nozzles (Hawkins & Hoch 1971; Pannu & Johannesen 1976). In recent years several researchers have tried to manipulate the internal contour of jet nozzles cleverly for thrust vectoring, enhanced mixing, and noise reduction. In the published literature such nozzles have been referred to as ‘asymmetric’ (Wlezien & Kibens 1988; Norum 1983), ‘scarfed’ (Lilley 1986), and ‘bevelled’ (Rice & Raman 1993*a,b*; Rice 1995). However, despite their benefits, such altered nozzles could screech differently, a concern that this paper addresses.

1.2. *Brief review of recent advances in understanding jet screech*

It is now well recognized that in imperfectly expanded jets, the screech tone is generated by coherent disturbances in the jets interacting with the shocks. The tone then propagates upstream (as feedback) to the jet exit and excites instabilities in the jet, thus closing the resonant loop. Pioneering work on screech was done by Powell (1953*a,b*), followed by investigations by Lassiter & Hubbard (1954) and Davies & Oldfield (1962*a,b*). Only a few of the many references on jet screech will be cited, but an extensive list of references is available in review articles by Tam (1991, 1995). According to Professor Tam, screech is the least understood component of supersonic jet noise, and details of the screech generation process and the physics of mode changes (in circular jets) remain unknown (see Powell, Umeda & Ishii 1992; Panda 1995). In addition, though models can predict the screech tone frequency (Tam 1988), no model predicts the amplitude or directivity of screech without recourse to empiricism. Screech, which is a special case of shock-associated noise (Harper-Bourne & Fisher 1974; Howe & Ffowcs Williams 1978; Tam, Seiner & Yu 1986), can also become intermittent and irregular. This was observed by Davies & Oldfield (1962*a,b*), but the reason for the intermittency remains unknown. Considerable progress toward understanding these issues has been made in the recent detailed studies on rectangular jets by Umeda & Yasuda (1990), Raman & Rice (1994), Walker, Gordeyev & Thomas (1995), Nishijima & Kaji (1995), and Cain *et al.* (1995). However, despite this progress, many aspects of the above issues remain unresolved.

1.3. *Objectives of the present work*

Key to predicting screech is modelling the shock-cell structure of supersonic jets. Researchers believe that changes in screech mode, amplitude, unsteadiness, and screech source structure are related to variations in shock-cell structures. In this connection considerable progress has been made in developing simple yet fairly accurate first-order estimates of the gross features of shock cells and screech tone frequencies of supersonic

jets (Tam, Jackson & Seiner 1985; Tam 1988; Morris 1988; Morris, Bhat & Chen 1989). The present work provides some clues to the relationship between screech mode changes and the shock-cell structures by intentionally producing spanwise variations in the shock cells and by studying changes in the spanwise screech mode. A rectangular geometry (aspect ratio = 5) was chosen, where the screech modes are always antisymmetric in the smaller nozzle dimension, allowing us to focus on mode changes in the spanwise direction.

The present work, which studies screech tones from rectangular jets with spanwise-non-uniform exits, considers three geometries: (a) a nozzle with a uniform straight edge, (b) a nozzle with a spanwise-oblique (single-bevelled) edge, and (c) a nozzle that has two equal and opposite spanwise-oblique (double-bevelled) edges that meet in the centre to form an arrowhead-shaped nozzle. Throughout the text, the three nozzle types will be referred to by their respective letters. For all nozzles considered, the transverse dimension had straight sides. The non-uniform geometries produced spanwise-oblique shock cells and spanwise instability modes. The modification of the shock cells and instability modes altered the spanwise screech modes, the screech frequencies, and the screech amplitudes. The three types of spanwise modes observed – symmetric (I), antisymmetric (II), and oblique (III) – will be referred to by their respective Roman numerals. Although II is a special case of III, this distinction will be maintained throughout the paper. Under conditions where more than one screech mode was present, this study also addresses the issue of whether the modes *coexisted* or were *mutually exclusive*.

1.4. Organization of the paper

The paper begins with a description of rectangular supersonic jet nozzles with spanwise oblique geometries (§2.1), followed by a description of the measurement and data analysis techniques (§2.2). The general characteristics of rectangular jets with non-uniform exits are described in §3.1, followed by a characterization of the spanwise-varying screech modes (§3.2). The relationship between spanwise screech modes and spanwise variations in the shock cell structure is described in §3.3. Screech unsteadiness and mode switching are discussed in §3.4, followed by a discussion of the screech source structure, strength, and location in §3.5. Finally the screech signature on a plane where damage is likely to occur from sonic fatigue is described in §3.6.

2. Experimental apparatus and procedure

2.1. Jet facility

The experiments were carried out at the NASA Lewis Research Center Jet Facility. Since the facility is discussed by Raman & Taghavi (1996), only a brief description is given here. The 76 cm diameter plenum tank was supplied with compressed air at pressures up to 875 kPa (125 Psig) at 26.7 °C (80 °F). After passing through a filter that removed dirt or dust, air entered the plenum axially where it was laterally distributed by a perforated plate and a screen. Two circumferential splitter rings that contained acoustic treatment (kevlar) removed upstream valve noise. The flow was further conditioned by two 50-mesh screens before exiting into the room through the nozzle. An automatic feedback control system maintained constant air-supply conditions. The control system could restrict pressure variations during each run to within 0.2%. Such precise control was essential for this experiment since the screech tone was extremely sensitive to changes in operating conditions.

The three nozzle types (a–c) are depicted with dimensions (cm) in figure 1 (a–c). The

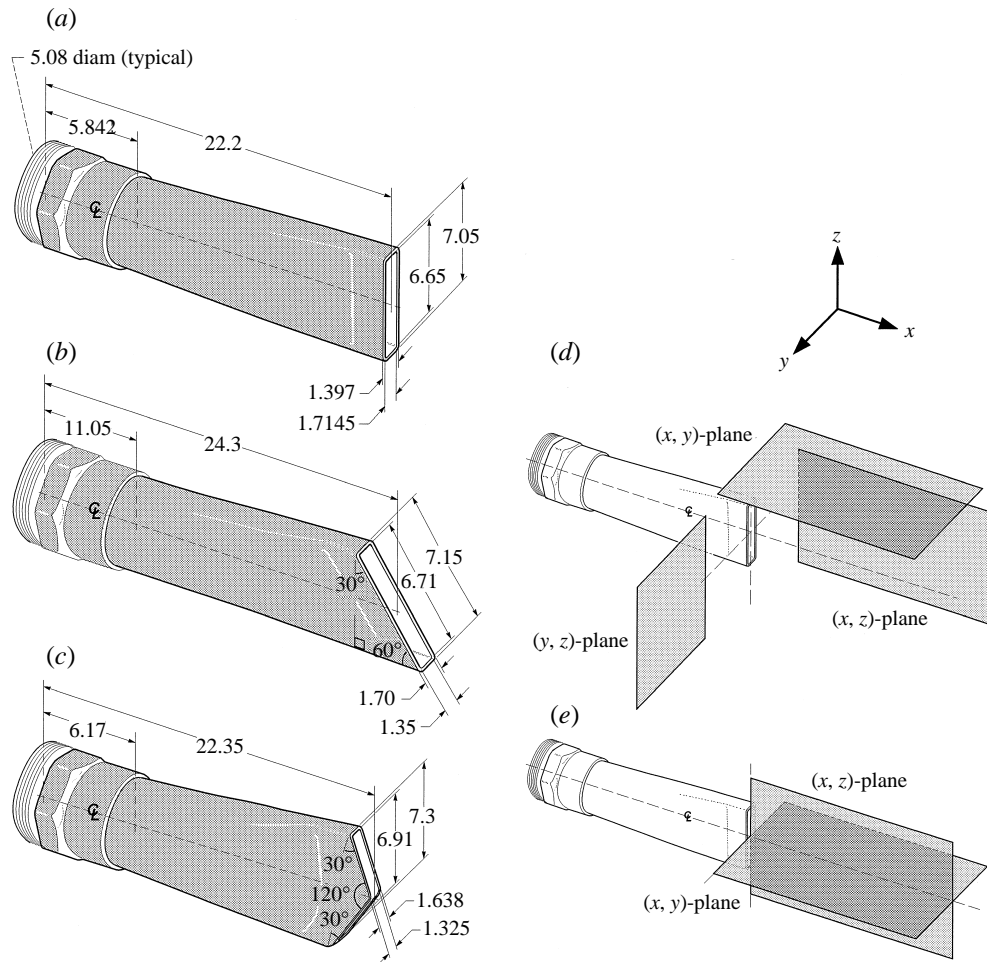


FIGURE 1. Nozzles and measurement planes (dimensions are in centimetres). (a) Spanwise-uniform nozzle, (b) single-bevelled nozzle, (c) double-bevelled nozzle, (d) noise measurement planes, (e) static pressure measurement planes.

nozzles were developed by Rice and used previously in the jet mixing and noise control studies of Rice & Raman (1993*a, b*), and Rice (1995). Each nozzle included a circular-to-rectangular transition section and a converging-diverging nozzle contour, all integrated into one piece. The area ratio of the convergence, $A_{circular}/A^*$, was 2.38. The divergence area ratio, A_e/A^* , was 1.128. Note that the subscript *e* and superscript * refer to conditions at the nozzle exit and throat respectively. Also note that the convergence-divergence occurred only in one direction (*y*), with straight sidewalls. As shown in figure 1 (*b, c*) all bevel cuts were made at 30° from the nozzle lip. The throat dimension for the three nozzles was 1.25, 1.17 and 1.17 cm for cases (*a*), (*b*) and (*c*) respectively. The nozzles, the probe traversing mechanism, and other reflective surfaces in the near field were covered with two layers of acoustically absorbent open-cell polyurethane foam (0.635 cm thick uncompressed). The idea was to minimize strong reflections from the nozzles and plenum. This material is known to be very effective in absorbing incident sound in the frequency range from 1000–25000 Hz (with several layers, lower frequencies can also be absorbed).

2.2. Measurement techniques

A spark schlieren system was used for flow visualization. The system included a Palfash light source, a microscope objective, two spherical mirrors (15.24 cm diameter, 91.44 cm focal length), and a vertical knife-edge. The light source consisted of an electric-arc in an inert atmosphere of argon gas, and could produce a 1 μ s pulse of high-intensity light (4 Joules). Photographs were taken by allowing light from the knife-edge to fall directly on Polaroid film.

The acoustic measurements were made in the near field using 0.64 cm (1/4 in.) diameter B & K microphones mounted under each nozzle and on a three-dimensional traversing mechanism for the near-field noise surveys. The noise measurement planes are shown in figure 1(d). The B & K microphones were omnidirectional within ± 1 dB up to 10 kHz and within ± 3 dB up to 20 kHz. The microphones were calibrated using a B & K pistonphone calibrator, with corrections for day-to-day changes in atmospheric pressure. The sound pressure levels reported in this paper are in dB relative to 20 μ Pa. The acoustic data were recorded using a B & K analyzer, and transient events were captured using a Spectral Dynamics instrument.

A short dual-cone static pressure probe designed by Pinckney (1975) at NASA Langley and used previously by Norum & Seiner (1982) was employed for the static pressure measurements in the jet. The static pressure measurement planes are shown in figure 1(e). Because the static pressure rises sharply downstream of a shock, the static pressure probe could be used to map the spanwise variations in the shock structure. The short static probe was less obtrusive than a longer version of this probe. However, when the short probe was calibrated against the more accurate longer version of this probe in a supersonic wind tunnel, the short probe was in error by as much as 7% (J. Panda, private communication). Additional errors were caused by flow angularity downstream of the oblique shocks. Despite these errors, the static pressure maps do allow a qualitative comparison of the shock-cell structures for the cases under consideration.

3. Discussion of results

3.1. General characteristics of jets from rectangular nozzles with non-uniform exits

Spark schlieren photographs of both the transverse and spanwise dimensions of the jet are shown for the three types of rectangular nozzles in figures 2–4. The photographs depict flows that are overexpanded ($M_j = 1.2$), perfectly expanded ($M_j = 1.4$), and underexpanded ($M_j = 1.6$), where M_j represents the fully expanded jet Mach number. For the nozzle with a straight exit (figure 2), the shock strengths are minimized at the design point ($M_j = 1.4$) and weak Mach waves are visible. At $M_j = 1.4$ the flow does not diverge significantly in the spanwise direction for the non-uniform nozzles (figures 3 and 4) because the spanwise pressure gradient cannot exert influence in a convergent–divergent (C–D) nozzle operated at its design point. However, due to difficulties in designing complex C–D nozzles, the flow from the bevelled nozzles contained some shocks even at the design Mach number. The spanwise asymmetry in the shock-cell structures is clearly seen in figures 3 and 4.

The focus of this paper is the flow from imperfectly expanded C–D nozzles. Two factors influenced the choice of C–D nozzles: first, such nozzles allow a study of overexpanded, design, and underexpanded conditions; second, when operated slightly off-design they have weak shocks that can be successfully modelled by linear shock-cell

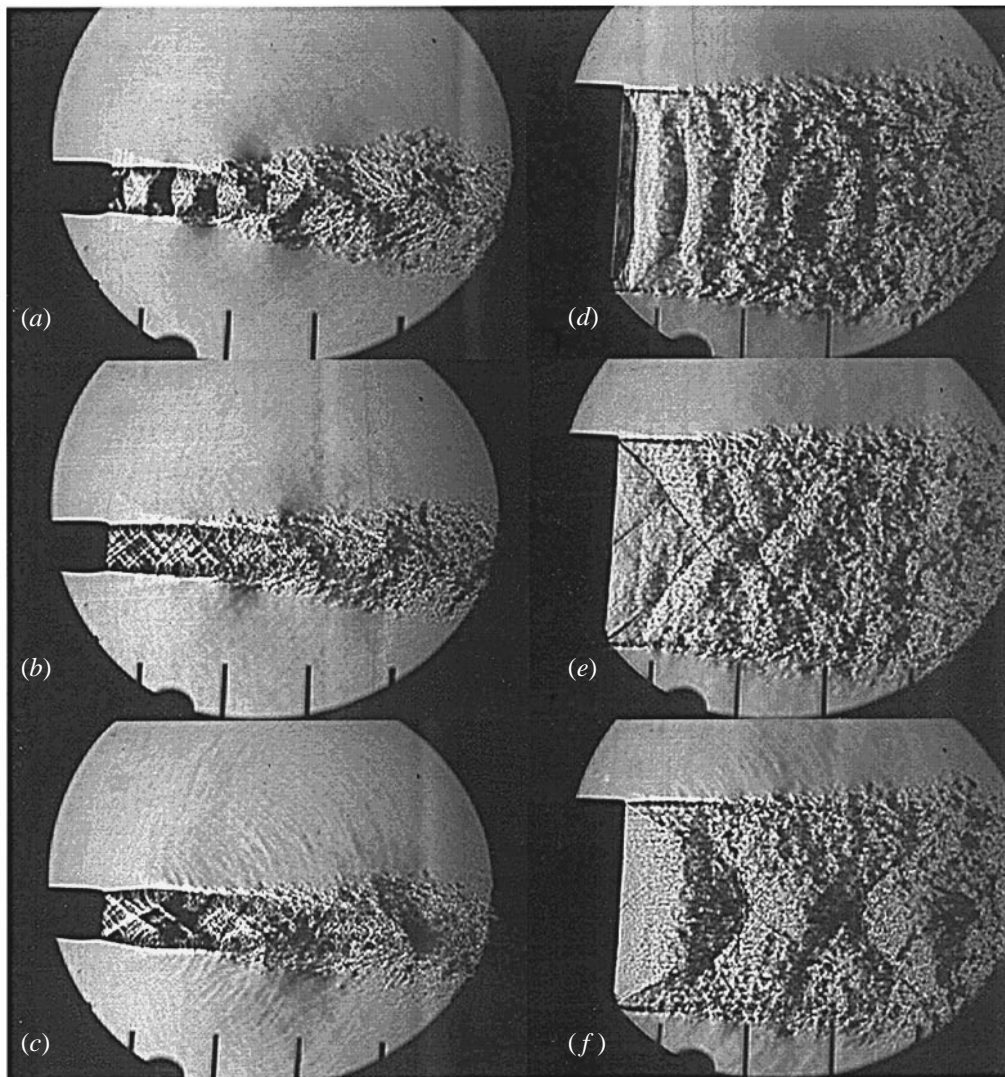


FIGURE 2. Spark schlieren images of over-, ideally-, and under-expanded jets from a rectangular convergent-divergent nozzle with a straight exit (both narrow and wide dimensions of the nozzle are shown). (a, d) $M_j = 1.2$, (b, e) 1.4, (c, f) 1.6.

models (Howe & Ffowcs Williams 1978; Tam *et al.* 1985; Tam 1988; Morris *et al.* 1989).

3.2. Characterization of spanwise-varying screech modes

The previous section described nozzle cases (a-c); screech characteristics for these nozzles are presented in figures 5-7. Information on the frequency, amplitude, and mode is presented in these figures. The Mach number axis is represented both as M_j and as $|M_a^2 - M_j^2|$, where M_a and M_j represent the design and fully expanded Mach numbers, respectively. The latter representation of the Mach number axis emphasizes the design point and allows comparisons between various degrees of over- and under-expansion. The data in figures 5-7 were obtained using three microphones (see the sketch in figure 5c) upstream of the nozzle exit. A pair of microphones measured the transverse phase difference ($\Delta\phi_1$) and another the spanwise phase difference ($\Delta\phi_2$). A

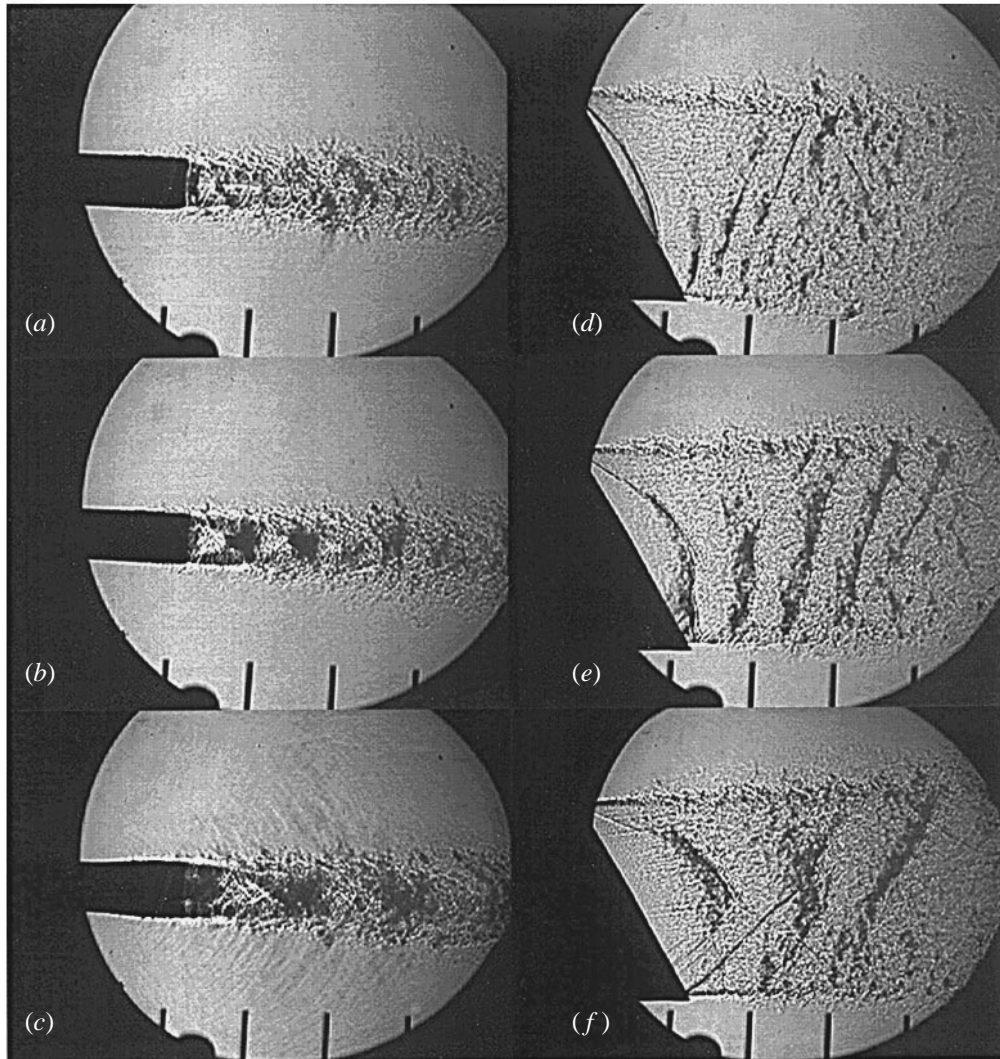


FIGURE 3. Spark schlieren images of over-, ideally-, and under-expanded jets from a rectangular convergent-divergent nozzle with a single-bevelled exit (both narrow and wide dimensions of the nozzle are shown). (a, d) $M_j = 1.2$, (b, e) 1.4, (c, f) 1.6.

common microphone was used as the reference microphone. All microphones were upstream ($x/h = -1.2$) of the nozzle exit, and the spanwise and transverse microphone spacings were $\Delta z = 2.5h$, and $\Delta y = 1.5h$.

The key to understanding the frequency and amplitude information presented in figures 5–7 is the phase information in part (c) of each figure. The transverse phase difference ($\Delta\phi_1$) was $+180^\circ$ (or -180°) for all cases, indicating that the screech mode was antisymmetric in the transverse direction for all nozzles over the entire Mach number range. Therefore, the distinction between modes will be based only on their spanwise variations. For the spanwise phase difference ($\Delta\phi_2$) there were three possibilities: 0° which signified a spanwise symmetric mode (type I); $+180^\circ$ (or -180°) which indicated the mode was spanwise antisymmetric (type II); non-zero and not 180° which indicated the screech mode was spanwise oblique (type III).

For nozzle (a) (figure 5), the following comments are warranted. The symmetric

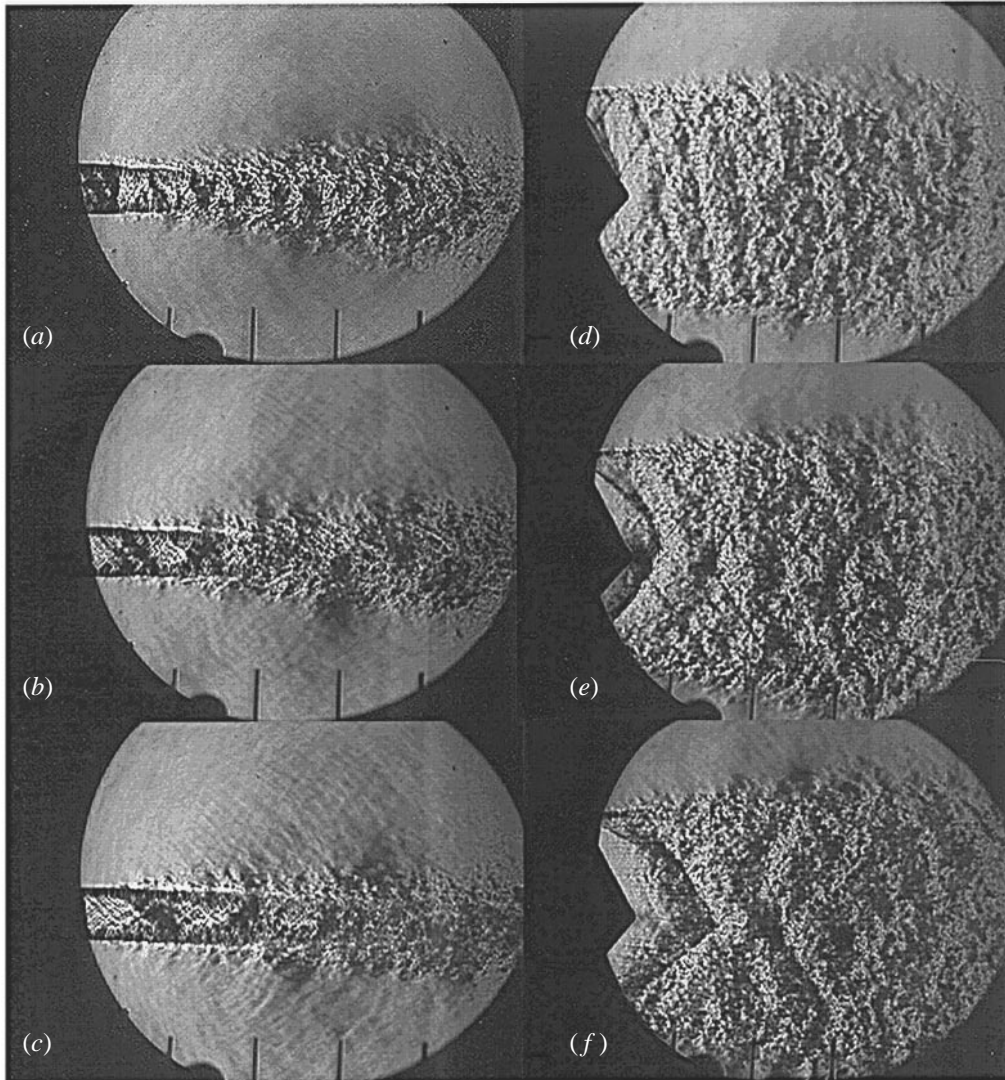


FIGURE 4. Spark schlieren images of over-, ideally-, and under-expanded jets from a rectangular convergent-divergent nozzle with a double-bevelled exit (both narrow and wide dimensions of the nozzle are shown). (a, d) $M_j = 1.2$, (b, e) 1.4, (c, f) 1.6.

mode (I) was present from $M_j = 1.25$ to 1.82. The corresponding Strouhal number range ($St(h) = fh/U_j$, where h is the smaller nozzle dimension and U_j is the fully expanded jet velocity) based on the frequencies in figure 5(a) was 0.28 to 0.06. Over the M_j range from 1.263 to 1.41 ($St(h) = 0.28$ to 0.2), the screech mode (I) weakened in amplitude (see figure 5(b)) and a spanwise antisymmetric mode (II) appeared. The new result here is that even rectangular jets with straight exits can produce spanwise antisymmetric screech modes when the shock-cell structure is weakened near the nozzle design point. This observation has either gone unnoticed or has been ignored by previous researchers.

For the single bevelled nozzle, (b) (figure 6), there is a single spanwise oblique mode (IIIA) between $M_j = 1.27$ and 1.6 ($St(h) = 0.209$ –0.095). Another type-IIIB mode appears at a different frequency between $M_j = 1.38$ and 1.53 ($St(h) = 0.183$ –0.135).

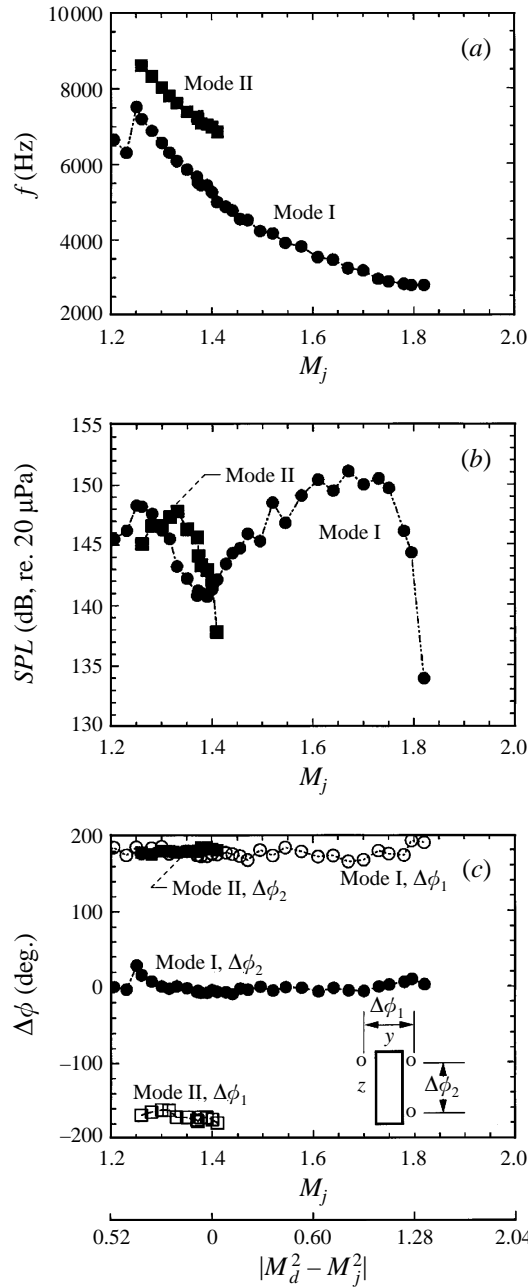


FIGURE 5. Screech tone characteristics of a rectangular convergent–divergent nozzle with a straight exit. (a) Screech frequency, (b) screech sound pressure level measured at the jet lip, (c) phase difference in the transverse and spanwise directions, versus the fully expanded jet Mach number.

Note from figure 6(b) that IIIA is diminished when IIIB is at its maximum amplitude. Beyond $M_j = 1.62$, both of the above modes disappear and a spanwise symmetric mode (I) appears. This type-I mode is sustained from $M_j = 1.62$ to 1.75 ($St(h) = 0.095$ to 0.07).

For the double bevelled nozzle, (c) (figure 7), below $M_j = 1.42$ there is no screech. A spanwise antisymmetric mode (II) appears between $M_j = 1.42$ and 1.66

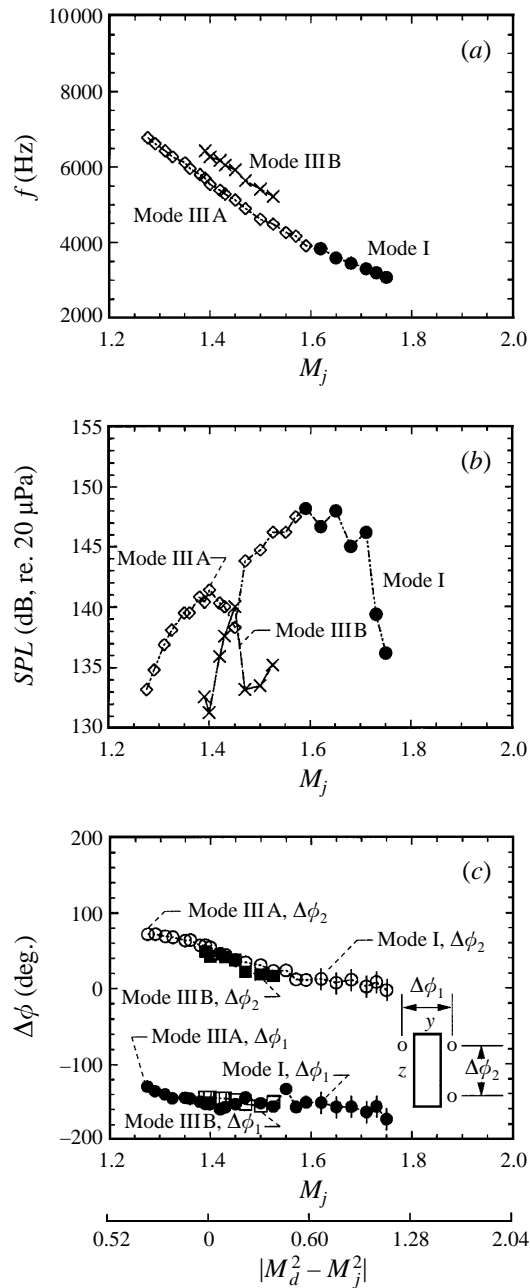


FIGURE 6. Screech tone characteristics of a rectangular convergent–divergent nozzle with a single-bevelled exit. (a) Screech frequency, (b) screech sound pressure level measured at the jet lip, (c) phase difference in the transverse and spanwise directions, versus the fully expanded jet Mach number.

($St(h) = 0.228$ to 0.139). For nozzle (c), it appears that the antisymmetric mode is formed by the superposition of two equal but opposite oblique waves produced by the oblique edges of the nozzle. Beyond $M_j = 1.66$, a mode switch occurs, and the screech mode becomes spanwise symmetric (I). Note that there is a reduction in screech amplitude (figure 7b) at the mode-switch point. The type-I mode is sustained from $M_j = 1.66$ to 1.87 ($St(h) = 0.139$ to 0.097). At high Mach numbers (approximately

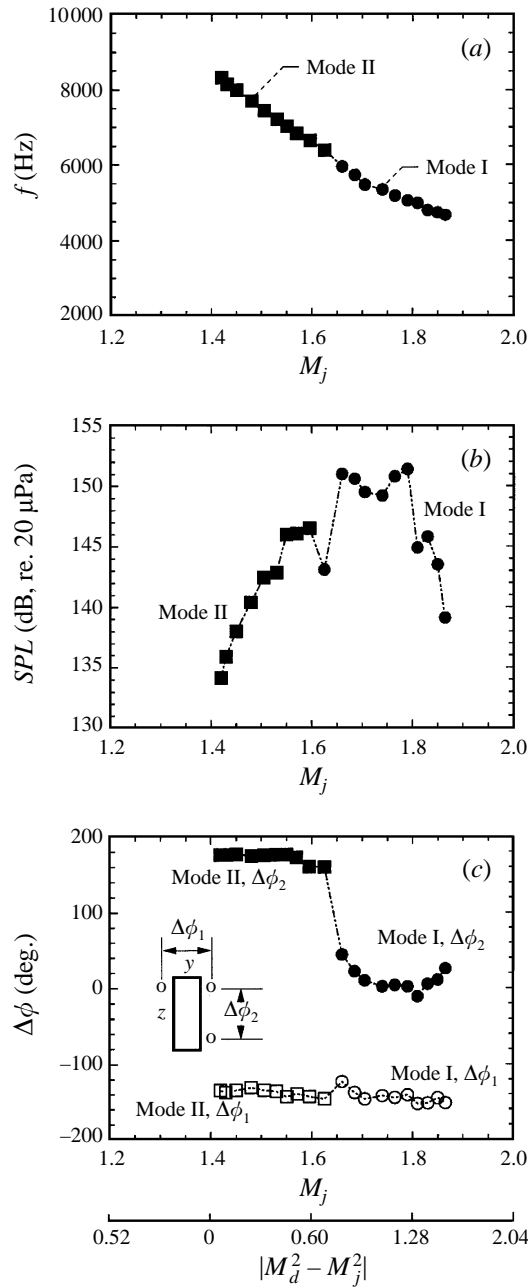


FIGURE 7. Screech tone characteristics of a rectangular convergent–divergent nozzle with a double-bevelled exit. (a) Screech frequency, (b) screech sound pressure level measured at the jet lip, (c) phase difference in the transverse and spanwise directions, versus the fully expanded jet Mach number.

beyond $M_j = 1.6$), the first shock turns into a normal shock (Mach disk). For all nozzles after the Mach disk forms, the screech mode becomes spanwise uniform.

For rectangular jets with uniform exits, the screech frequency (f) can be predicted if the shock-cell spacing (λ) is known (Tam 1988; Morris *et al.* 1989). In such predictable cases the first four shock cells are uniform and equally spaced. However, in the present case, owing to spanwise variations and irregularities in the shock-cell

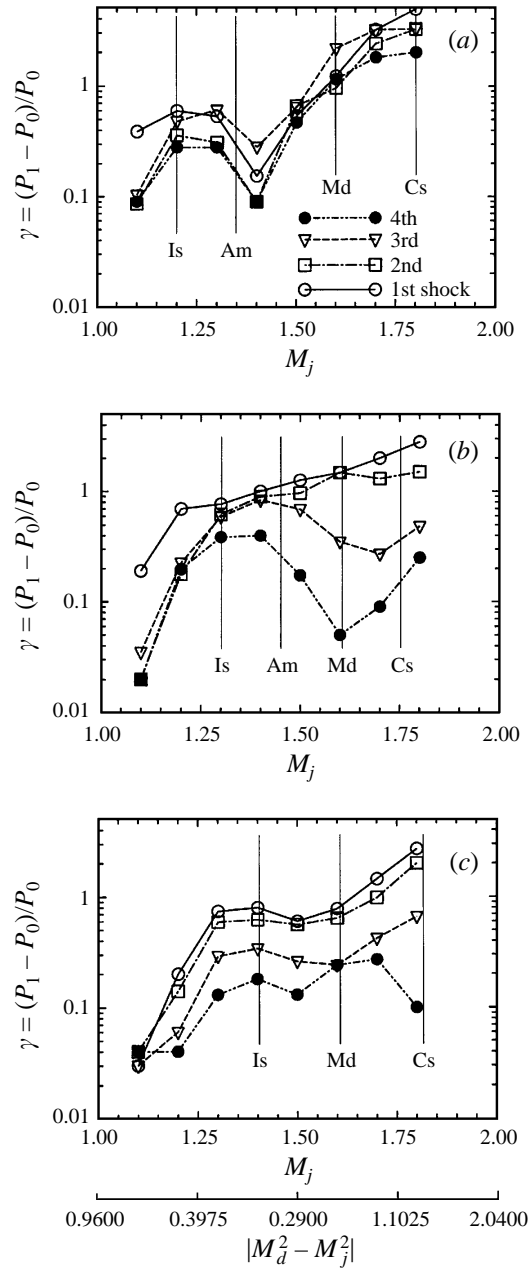


FIGURE 8. Shock strength versus fully expanded jet Mach number. (a) Rectangular nozzle with straight exit, (b) single-bevelled nozzle, (c) double-bevelled nozzle. (Is – initiation of screech, Am – auxiliary mode appears, Md – Mach disk forms, Cs – cessation of screech).

structure, and the presence of multi-modes, the centreline average shock spacing is not a very meaningful indicator. A much more elaborate description (see shock patterns in figures 2–4) is required to develop models that can predict the screech frequencies and modes that were measured in the present experiment.

The strengths of shocks 1–4 (measured on the jet’s centreline) versus Mach number are given in figure 8(a–c) for the three cases under consideration. The shock strengths

are given by $\gamma = (P_1 - P_0)/P_0$ where P_1 and P_0 are the static pressures downstream and upstream of a shock. In the present argument the strengths of shocks 1–4 demonstrate the relationship between shock strength and screech tones. Although some researchers believe that the connection between screech and shocks must be a complex phenomenon, the present data demonstrate clearly that a fairly simple but significant notion prevails: screech initiation and destabilization depend on shock strength.

First, from figure 8(a–c) it is clear that the initiation of screech occurs only after a critical shock strength is reached. Second, when the shock structure is weakened (e.g. C–D nozzle operated at its design point), the main screech mode is destabilized and an auxiliary screech mode appears. Finally, the formation of the Mach disk makes all screech modes spanwise uniform, regardless of nozzle geometry. The disappearance of screech at high M_j (cessation) is not caused by decreasing shock strength: screech cessation is attributed to the diminished feedback and receptivity that occur at very high values of M_j (see Raman 1996; Cain & Bower 1996). An additional aspect revealed here is that, even within the M_j range where strong screech occurs, a weakening of shocks 3 and 4 can destabilize screech.

Note also that the argument presented in the previous paragraph does not contradict Powell's (1953*a, b*) idea that the strongest screech tones are generated when the phases of pressures radiated from each shock cell best reinforce each other at the nozzle exit. Although there are numerous sources, the effective source centre depends only on one or two shock cells (depending on M_j). Typically the 3rd or 4th shocks are the effective source centres, which is not difficult to comprehend because near the jet exit the shock strengths are very high, but the instabilities are still growing there. Far downstream the shock strengths diminish and the jet's coherent structure is also subject to dissipation. Consequently, the dominant acoustic centre is at shock 3 or 4 (reported to be at shock 3 by Krothapalli *et al.* 1986). When the source distribution displays a clear acoustic centre, the problem is similar to that of an edgetone (Powell 1961; Karamcheti *et al.* 1969; Crighton 1992). However, unlike the edgetone, the screech source location is a variable that depends on the jet's operating conditions and dimensions. The aforementioned trends are clearly borne out by the data. However, it is disappointing not to be able to make more definite quantitative predictions about screech using the shock strength values. The problem here is not the inadequacy of the experiment but our ignorance of the screech generation phenomenon. In a later section it will be shown that screech intermittency and peak source amplitude are dependent on the strengths of shocks 3 and 4.

3.3. Relationship between spanwise-varying screech modes and spanwise-oblique shocks

A mode map summarizing previously described modes for nozzles (a–c) is given in figure 9. We now direct attention to the connection between spanwise shock structure and screech modes. Static pressure maps depict the spanwise shock-cell structures for the three cases under consideration (see figures 10–12). Since the static pressure rises sharply downstream of a shock, details of the shock-cell structure are apparent from such maps. The measurement was made on an (x, z) -plane at $y/h = 0$ (see figure 1e). A typical measurement included 1071 data points (21×51) with $\Delta x/h = 0.15$, and $\Delta z/h = 0.23$. Similar measurements made on the (x, y) -plane are not shown here. If we compare the mode map of figure 9 with the static pressure maps (figures 10–12), it is evident that for all three nozzles under consideration the screech modes corresponded with the spanwise shock structure (i.e. spanwise-uniform shocks produce spanwise-uniform screech modes, and spanwise-non-uniform shocks produce spanwise-non-

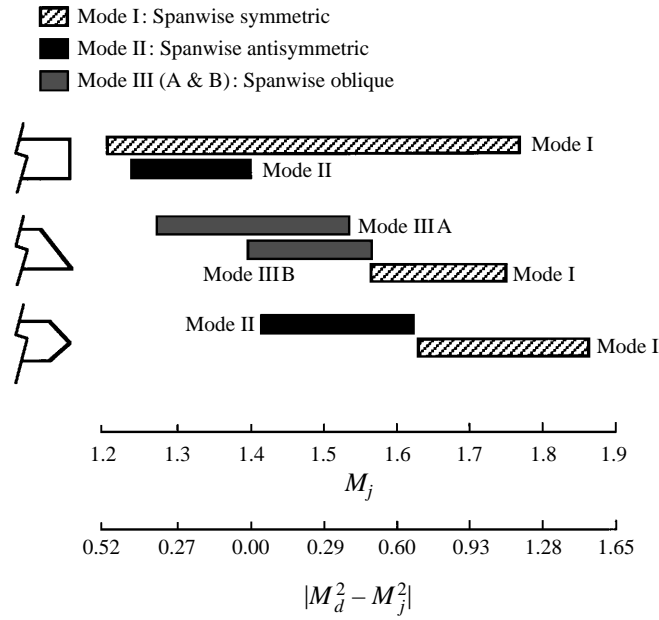


FIGURE 9. Mode diagram for rectangular jets with spanwise-oblique nozzle exit geometries.

uniform screech modes). It also needs to be emphasized that the shock-cell structure, not the nozzle, geometry, determined the spanwise screech mode. For example, the antisymmetric mode was produced by the double-bevelled nozzle and by the nozzle with a straight-edged nozzle whose dominant screech mode has been destabilized. The spanwise-oblique mode was produced by the nozzle with a spanwise-oblique exit geometry until the Mach disk formed. The spanwise-uniform mode was produced by the straight-edged nozzle and by all nozzles after the Mach disk had formed, since the structure of downstream shocks was not influenced by nozzle geometry. As mentioned earlier, after the Mach disk formed, all modes were spanwise-uniform regardless of nozzle geometry.

Regarding the screech mode change, two types of mode switch are evident from the data of figures 5–8 (the corresponding shock structure is depicted in figures 10–12). The first type occurs when the dominant screech mode is weakened (due to weakening of shocks) and an auxiliary screech mode appears. In §3.4 it will be shown that the main and auxiliary screech modes can *coexist* or be *mutually exclusive*. The second type of mode switch occurs when one mode ceases to exist and another takes over. This type of switch occurs only when the formation of the Mach disk changes all spanwise-non-uniform modes to spanwise-uniform modes. The former type of mode switch is seen clearly for nozzles A (mode I to II), and B (mode IIA to IIIB), and the latter type of mode switch is seen for nozzles B (mode III to I), and C (mode II to I). The above results suggest that a mode switch of the first type involves a switch in the effective screech source from one shock to another of a different structure, whereas a mode switch of the second type involves a change in the structure of the shock that plays the role of the effective source.

These results are significant for the general issue of screech mode change. For example, in a round jet, researchers are still puzzled about why and how mode staging occurs (Tam 1995; Powell *et al.* 1992), although its occurrence has been documented by several investigators. It is not clear whether mode changes with increasing Mach number are due to changes in the evolution of coherent structures or due to changes

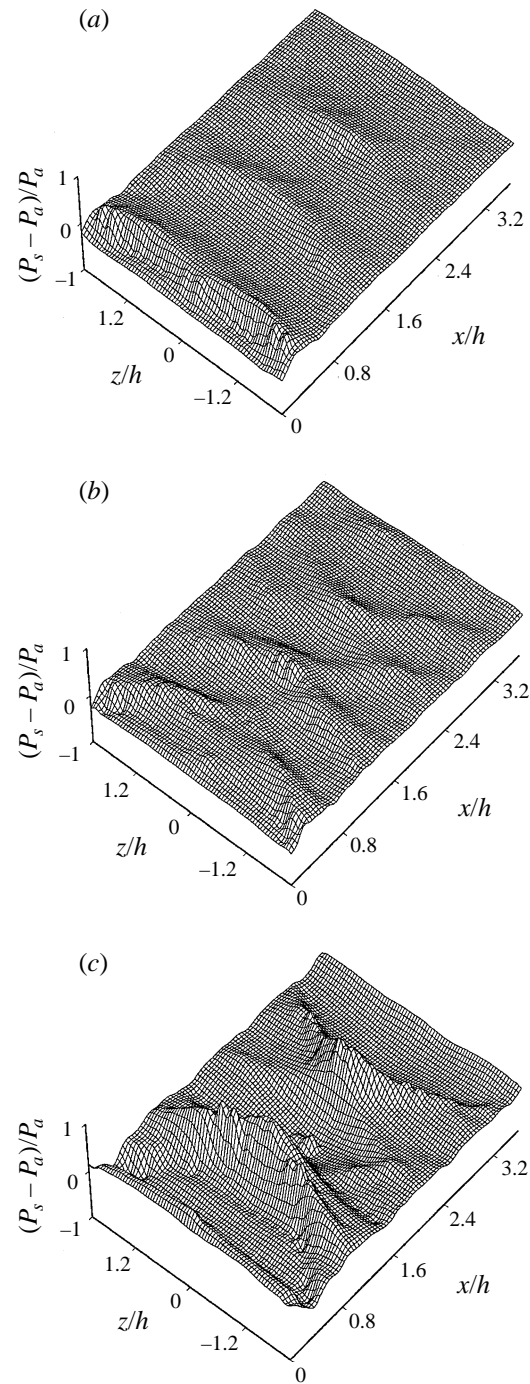


FIGURE 10. Spanwise variations in the shock-cell structure for a jet from a rectangular nozzle with a straight exit. The (x, z) -plane was located at $y/h = 0$. (a) $M_j = 1.2$, (b) 1.3, (c) 1.7.

in the shock-cell structure or whether one can even separate the two effects. In the present experiment the spanwise structure of the shock cells was deliberately changed to study the resulting screech mode change. The spanwise variations in the shock cells

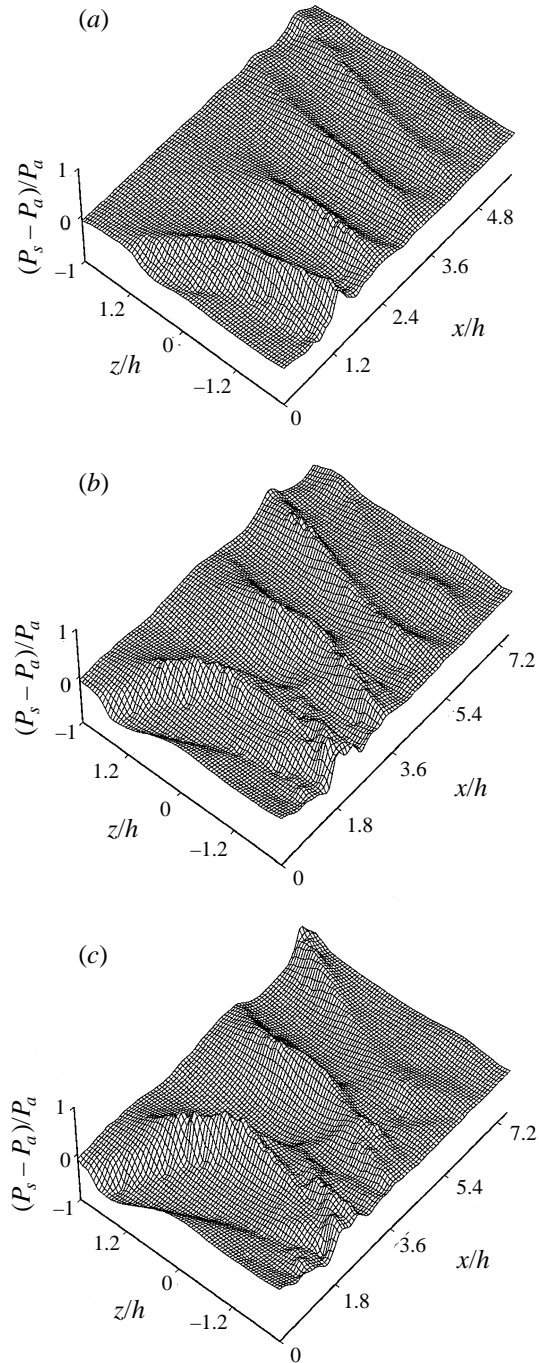


FIGURE 11. Spanwise variations in the shock-cell structure for a jet from a rectangular nozzle with a single-bevelled exit. The (x, z) -plane was located at $y/h = 0$. (a) $M_j = 1.3$, (b) 1.5, (c) 1.6.

do change the screech modes accordingly. Although the round jet problem is not within the scope of the present work, the cases described here do provide some clues.

3.4. Screech unsteadiness and mode switching

The mode map (figure 9) illustrated that at certain operating conditions more than one

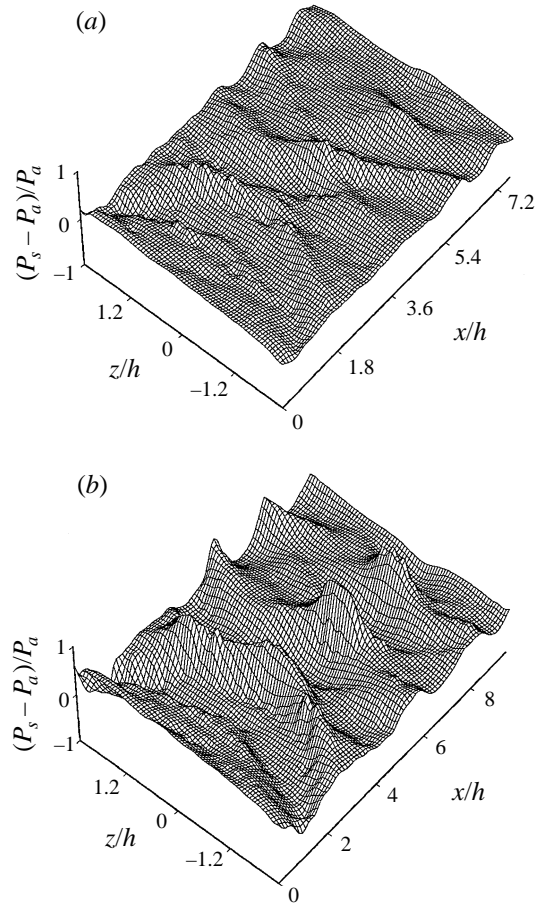


FIGURE 12. Spanwise variations in the shock-cell structure for a jet from a rectangular nozzle with a double-bevelled exit. The (x, z) -plane was located at $y/h = 0$. (a) $M_j = 1.5$, (b) 1.7.

screech mode can be present. The mode map was constructed using time-averaged data and therefore cannot reveal whether two modes are *coexisting* or switching in a *mutually exclusive* fashion. Walker *et al.* (1995) used the wavelet transform to address the time–frequency localization issue. In this paper the *intermittency* issue is addressed using ‘instantaneous spectra’. The ‘instantaneous spectra’ were obtained by performing a fast Fourier transform (FFT) of smaller segments (with a 40% overlap) of a long time sequence (4.816 s). The FFT block size as 1024 and the sampling rate was 25.6 KHz, providing spectral data in a range from 0 to 10 KHz with a frequency band width of 25 Hz. The result is displayed as a ‘waterfall’ plot of ‘instantaneous spectra’. Figures 13–15 show long-time-averaged spectra (0–25.6 KHz, 32 Hz band width) alongside ‘instantaneous spectra’ (0–10 KHz, 25 Hz band width) for all three nozzles over the Mach number range from 1.3 to 1.7. For the spanwise-uniform nozzle both modes (types I and II) coexisted at $M_j = 1.3$ and 1.4 (see figure 13), beyond which only a steady single dominant mode was present. For the single-bevelled nozzle, a weak screech mode (type IIIA) was present at $M_j = 1.3$ (see figure 14). As this mode gained strength at $M_j = 1.4$, another mode (type IIIB) appeared. At $M_j = 1.5$ both oblique modes (IIIA, B) competed for dominance almost in a mutually exclusive manner.

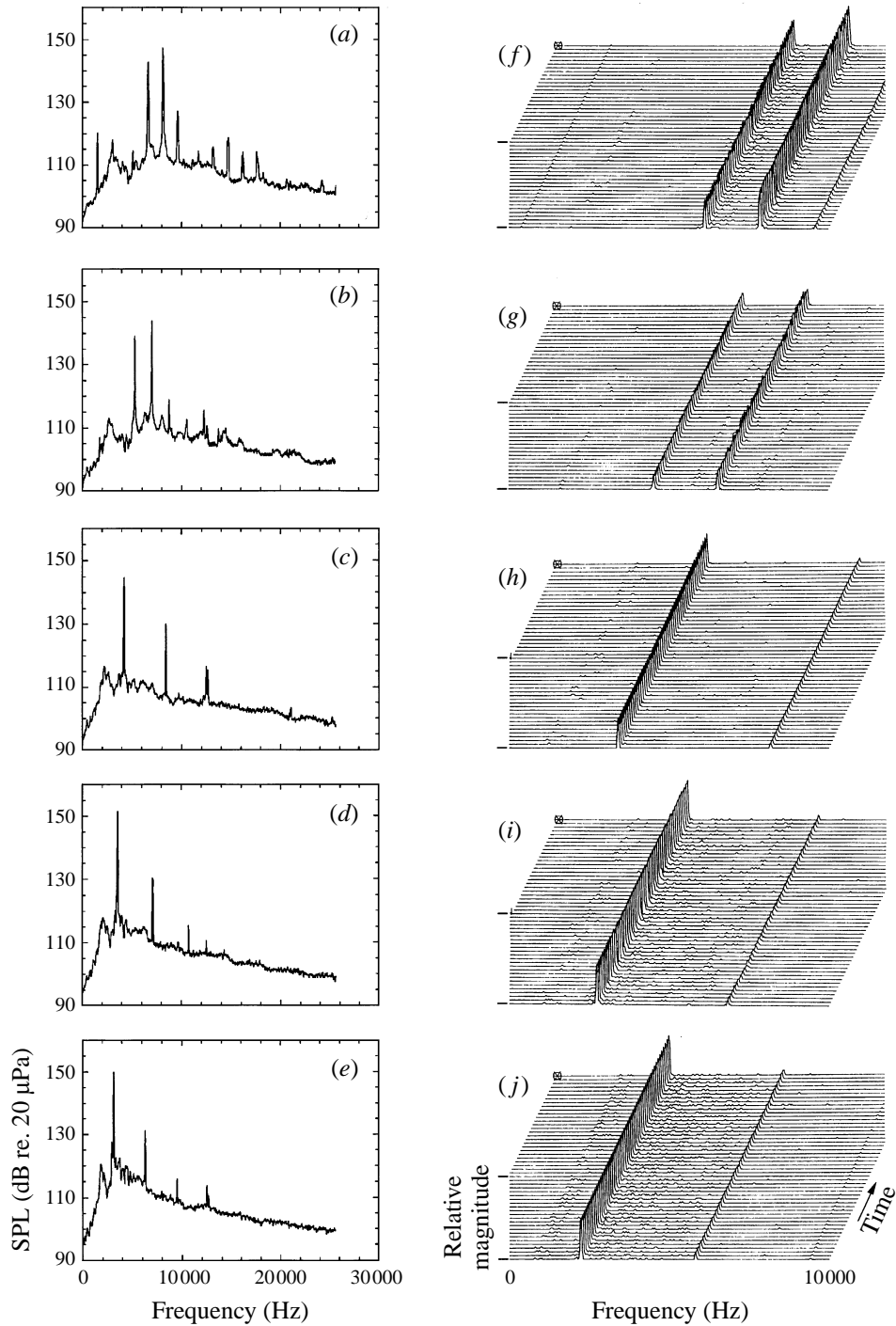


FIGURE 13. Evaluation of screech unsteadiness by contrasting time-averaged spectra with a waterfall plot of instantaneous spectra for a jet from a rectangular nozzle with a straight exit. (a,f) $M_j = 1.3$, (b,g) 1.4, (c,h) 1.5, (d,i) 1.6, (e,j) 1.7.

Beyond $M_j = 1.6$ only a single mode (type I) was present, and that became *irregular* at $M_j = 1.7$.

Finally, the double-bevelled nozzle displayed very weak screech modes below

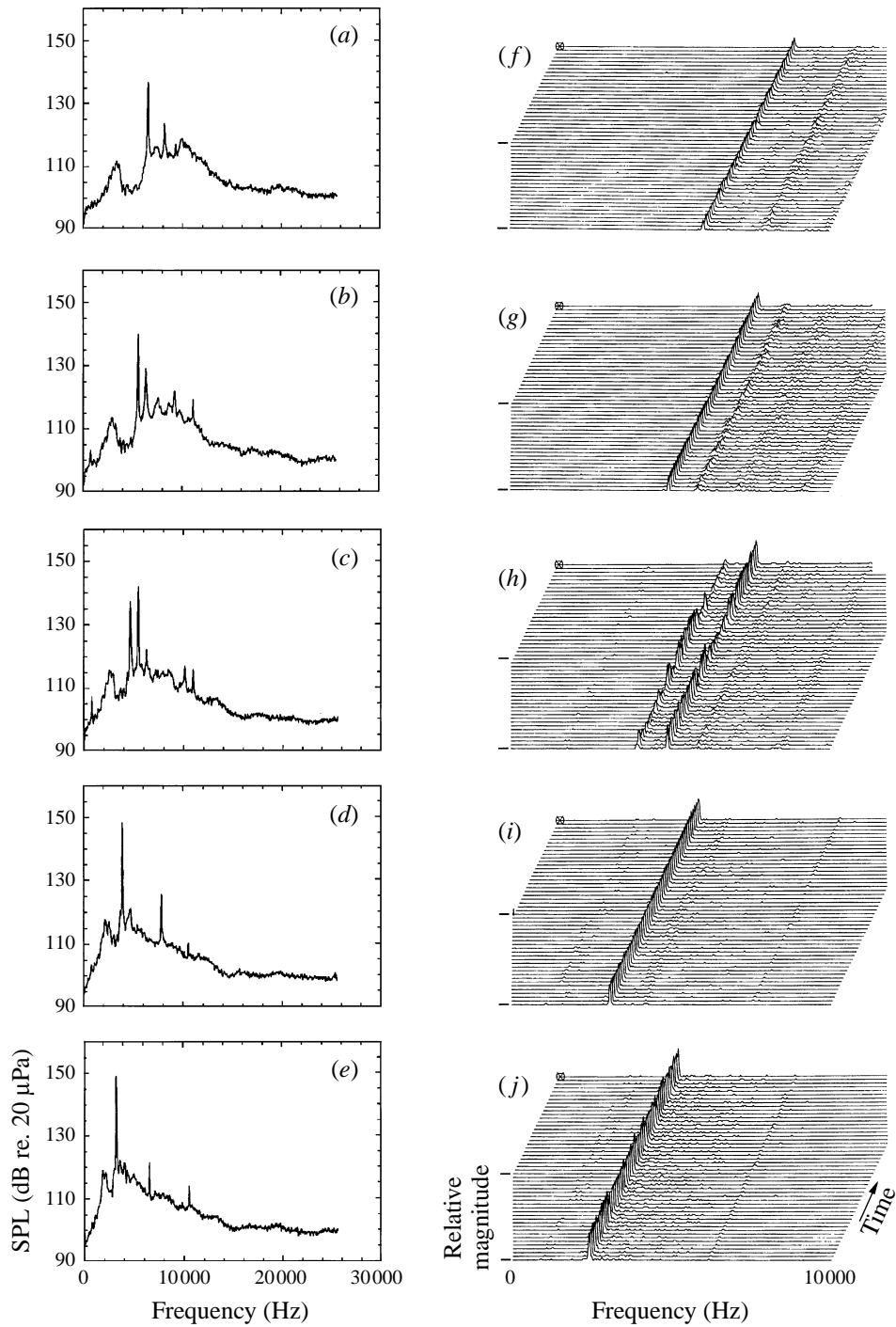


FIGURE 14. Evaluation of screech unsteadiness by contrasting time-averaged spectra with a waterfall plot of instantaneous spectra for a jet from a rectangular nozzle with a single-bevelled exit. (a,f) $M_j = 1.3$, (b,g) 1.4, (c,h) 1.5, (d,i) 1.6, (e,j) 1.7.

$M_j = 1.5$ (see figure 15). At $M_j = 1.5$, the antisymmetric (II) screech mode was steady. The mode switch (from II to I) occurred around $M_j = 1.6$, beyond which the spanwise-uniform mode was very intermittent and irregular.

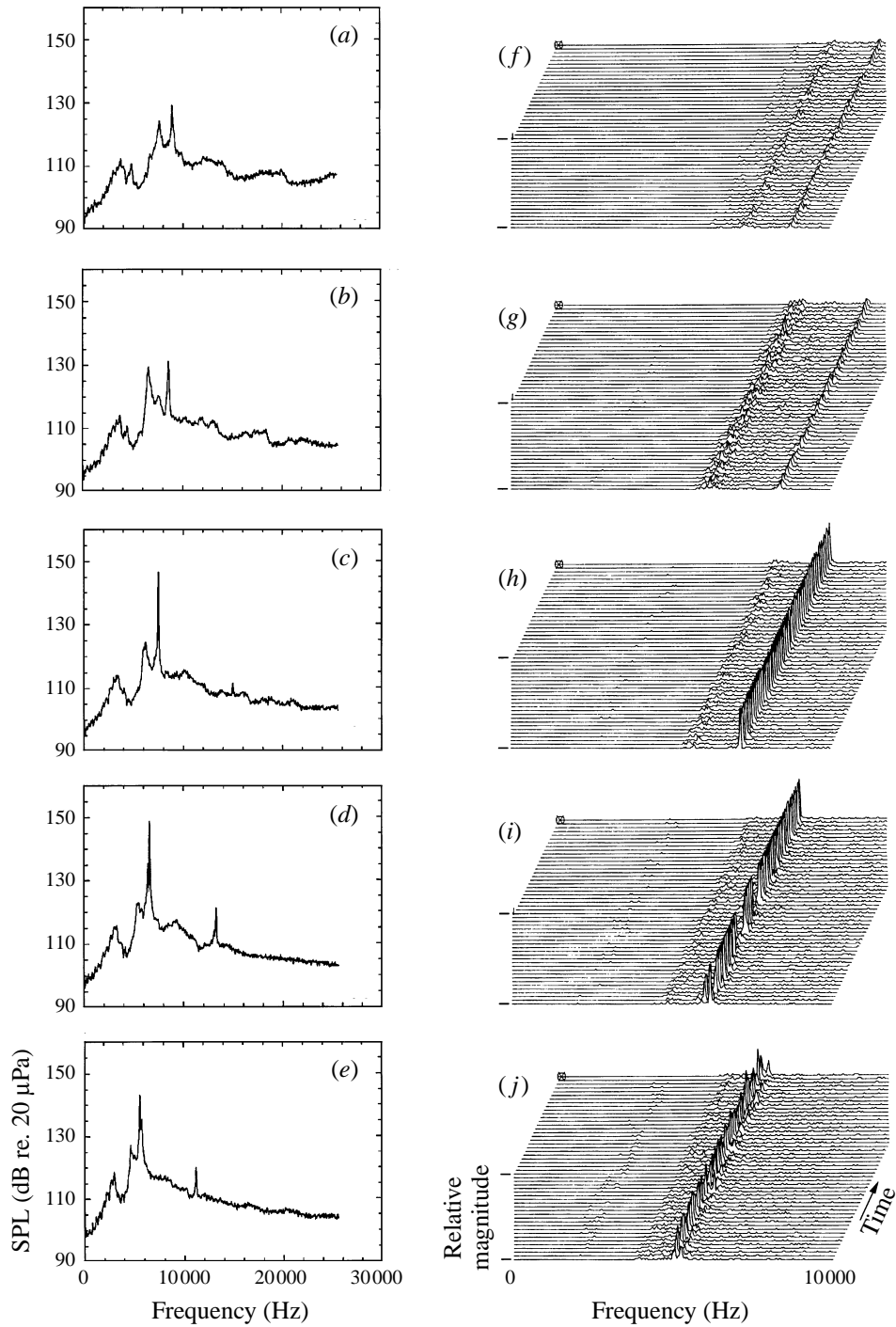


FIGURE 15. Evaluation of screech unsteadiness by contrasting time-averaged spectra with a waterfall plot of instantaneous spectra for a jet from a rectangular nozzle with a double-bevelled exit. (a,f) $M_j = 1.3$, (b,g) 1.4, (c,h) 1.5, (d,i) 1.6, (e,j) 1.7.

The reader is asked to refer back to figure 8 and observe that the amplitude and intermittency of screech are connected to the strengths of shocks 3 and 4. When the shock strength decreases, the screech amplitude diminishes. Evidence for the above

trend can be found by comparing figure 8(a) to figure 13. For the straight nozzle, at the design point ($M_j = 1.4$), the shock strength is minimized (figure 8a) resulting in a decrease in the screech amplitude (figure 13g) and the appearance of mode II as discussed earlier. For the single-bevelled nozzle the strengths of shocks 3 and 4 begin to diminish at $M_j = 1.5$ (figure 8b), and this coincides with unstable competing modes in figure 14(h). Finally for the double-bevelled nozzle, the strength of the 4th shock is reduced beyond $M_j = 1.6$, and the screech mode becomes irregular and intermittent (figure 15i,j). The above examples demonstrate that the strengths of shocks 3 and 4 are indicative of the amplitude and steadiness of screech and should be considered as a factor in screech calculations.

3.5. Screech source structure, strength, and location

In §§3.1–3.4 screech mode results for nozzles (a–c) over the entire range of Mach numbers were discussed; a single Mach number ($M_j = 1.5$) will now be described in detail. Contours for equal sound pressure levels at the screech frequency on the (x, z)-plane (see figure 1d) are shown in figure 16. Similar measurements made on the (x, y)-plane are not shown here. Contours of low sound pressure levels are shown as dashed lines since our focus is on the effective sources that are given by the high-amplitude islands. For the islands of contours under study, several observations can be made. For a nozzle with a straight exit (figure 16a), these islands appear to be centred on the $z/h = 0$ line, whereas for the single-bevelled nozzle (figure 16b, c) that produces oblique shocks, the high-amplitude islands for both oblique modes (III A, B) are off-centre and oblique in the spanwise direction. The most interesting case is that of the double-bevelled nozzle (figure 16d). Since this nozzle produces pairs of equal and opposite oblique shocks, two rows of high-amplitude islands on either side of the spanwise centreline appear. The peak amplitude at the apparent screech source is highest for the nozzle with the straight edge (166.6 dB). Corresponding peak levels for the single- and double-bevelled nozzles are 158.3 dB and 153.2 dB. At $M_j = 1.5$ the strengths ($\gamma = (P_1 - P_0)/P_0$) of shocks 3 and 4 for the three nozzles were 1.213 and 0.645, 0.699 and 0.174, and 0.259 and 0.133 for nozzles a, b, and c, respectively. Finally, it must be pointed out that the correspondence between shock strength and screech source amplitude is clear only when other factors such as instability wave growth, feedback, and receptivity are not dominant factors in deciding the fate of screech.

Our source location results are not affected by the refraction of sound as it emerges from the jet (for a description of sound refraction by flow see Morris, Richarz & Ribner 1973; Crighton 1975). The refraction issue was further clarified by Professor Morris in a private communication, according to which: first, refraction by mean velocity takes some wavelengths to be seen; second, there are no refraction effects due to mean velocity at 90° to the flow direction. It follows that the source locations depicted in figure 16 are accurate.

The relative phase corresponding to the amplitude information of figure 16 is shown in figure 17. The phase data are shown only for $x/h = 0$ to 4. Although phase data were acquired for x/h from 0 to 8 as in figure 16, credible phase data are available only for $x/h = 0$ to 4 for the following reasons. The phase difference was obtained from a cross-spectrum (phase) between the moving microphone and a reference microphone upstream of the nozzle ($x/h = -1.25$, $y/h = 0.75$, $z/h = 1.25$). The raw analysis provides data between -180° and $+180^\circ$. When one of these limits is exceeded there is an abrupt jump (readjustment) of the phase angle. In a simpler situation with continuous phase, it is easy to unwrap the phase and determine the proper quadrant for the phase data. In the present case there are actual jumps in phase near screech sources and across shocks. Thus in regions far downstream it is not always possible to

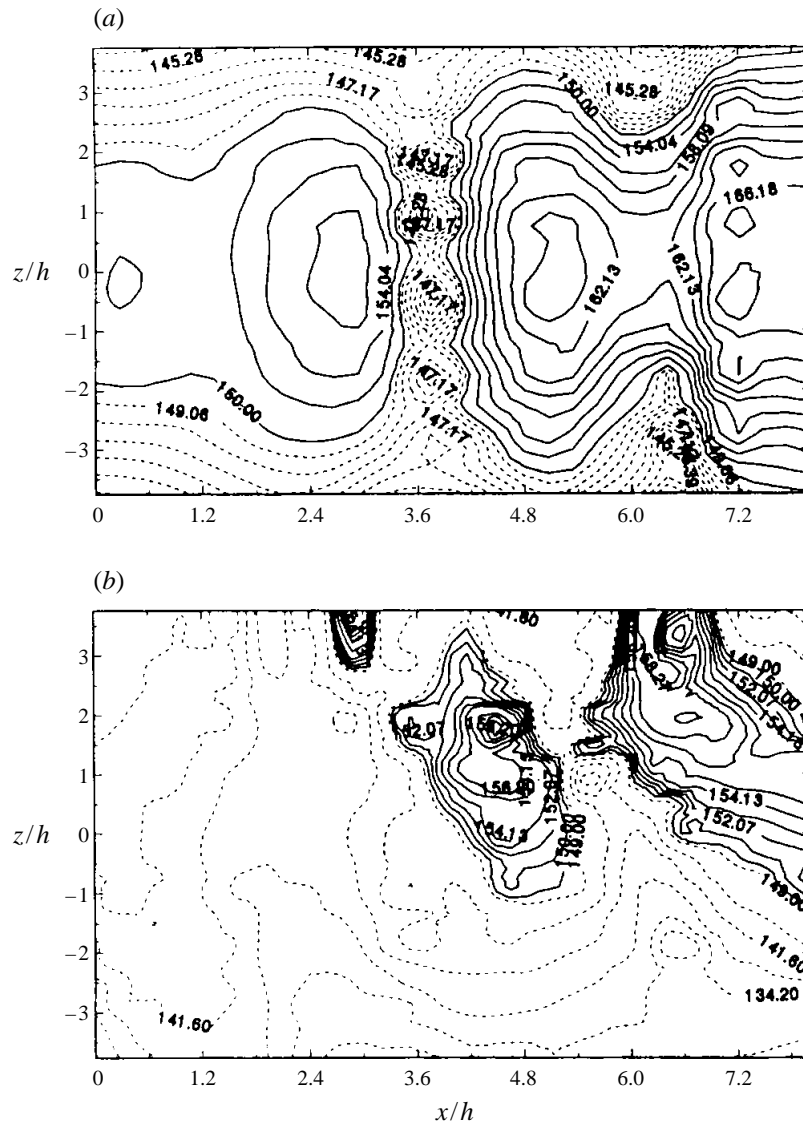


FIGURE 16 (a, b). For caption see facing page.

distinguish a physical phase jump from one that is dependent on the instrument (see Rice & Taghavi 1992). For this reason, data are shown only in regions where they are deemed credible. The isophase contours depict the spanwise screech mode (i.e. symmetric, oblique, and antisymmetric). Note that the inclination of the phase lines in figure 17(b–c) is not the bevel angle of the nozzle but the spanwise angular orientation of the screech source in figure 16(b, c).

The shock strength variations in the spanwise direction corresponding to the amplitude and phase information of figures 16 and 17 are shown in figure 18. Part (a) of figures 16–18 show that the source amplitude contours (figure 16a), phase lines (figure 17a) and shock cell (figure 18a) are all centred at and symmetric about $z/h = 0$. Likewise, spanwise-oblique sources (figure 16b, c), and oblique phase lines (figure 17b, c) correspond to oblique shock patterns (figure 18b, c). Finally, the twin sources

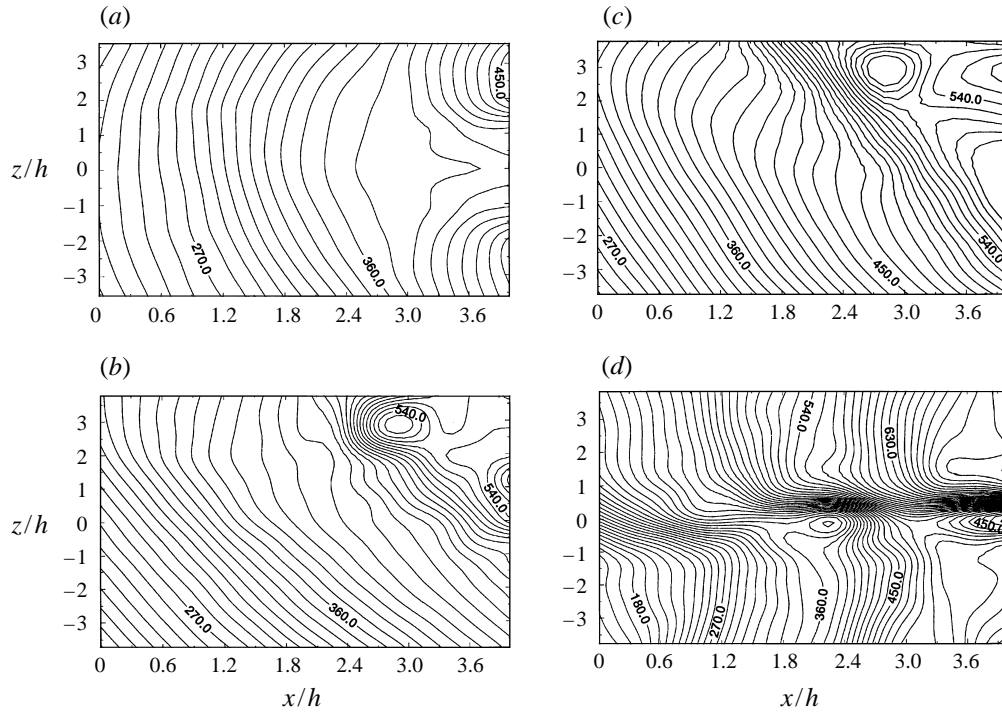


FIGURE 17. Isophase contours of spanwise-varying screech modes. The (x, z) -plane was located at $y/h = 1.5$, $M_j = 1.5$. (a) Rectangular nozzle with straight exit, (b, c) single-bevelled nozzle's dual mode, (d) double-bevelled nozzle.

introduction, one of the concerns is the propagation of screech upstream causing structural damage. The 'damage plane' is where structural damage is likely to occur from sonic fatigue (see figure 1d) – the (y, z) -plane just behind the nozzle exit.

For the data shown in figure 19 the (y, z) -plane was located at $x/h = -1.2$. The sound pressure level and phase map on this plane for the three cases under consideration reveal several interesting features. For both the uniform and the single-bevelled nozzles, the sound pressure level (SPL) peaks around $z/h = 0$, whereas for the double-bevelled nozzle the SPL exhibits a minimum at $z/h = 0$. For the latter nozzle the sound pressure level peaks are near the spanwise edges of the nozzle and are consistent with the source location shown in figure 16(c). The phase data reveal a 'null region' (where the phase does not change) for the nozzle with a straight edge. This null region was described in a previous paper (Raman & Taghavi 1996) and is produced at high Mach numbers because, as M_j increases the sources of screech move downstream, and therefore the wavefronts arriving at the nozzle exit plane would be flatter in the near-nozzle region. Figure 19(d–f) indicates the relative phase (with respect to a reference microphone) of a microphone that was moved in the (y, z) -plane. For the nozzle with a straight exit (figure 19d) the phase variation in the z/h -direction (at $y/h = 1$) is very small, because the screech mode is spanwise uniform. However, for the single-bevelled nozzle (figure 19e), at $y/h = 1$, the phase variation in the z/h -direction is 90° between $z/h = -4$ and $+4$, and is indicative of a spanwise-oblique mode. The spanwise phase difference is 180° for the case of the double-bevelled nozzle (figure 19f), indicating that the screech mode here is spanwise antisymmetric. Thus, the phase for

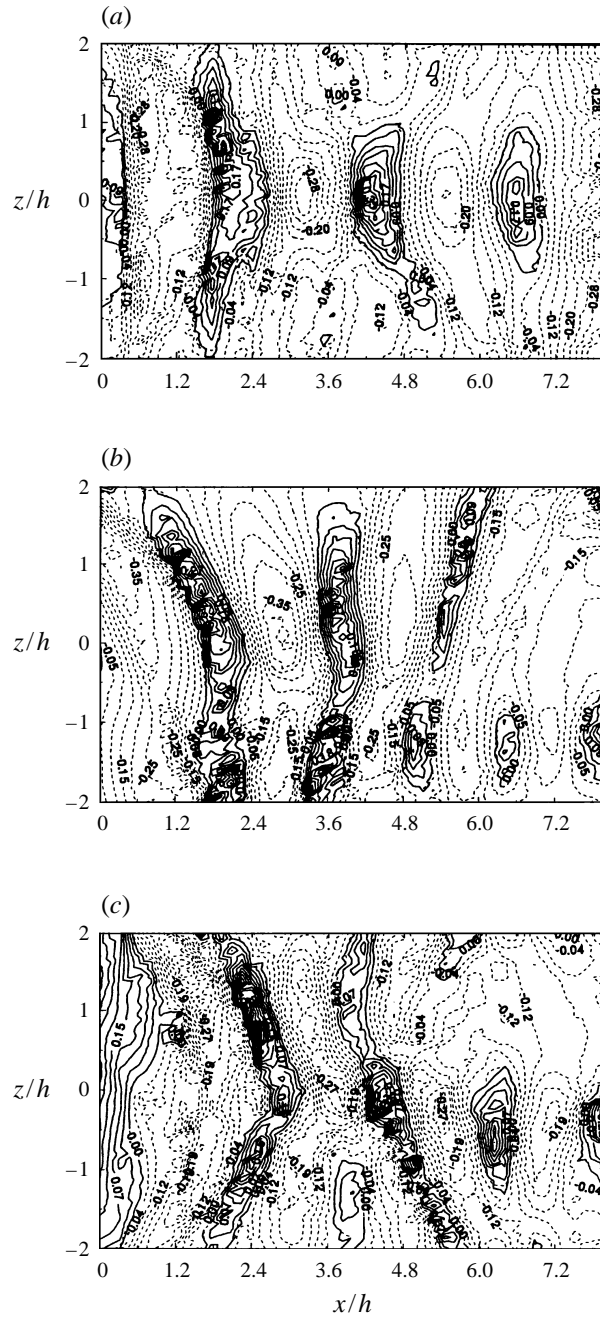


FIGURE 18. Spanwise shock-cell structure corresponding to the screech amplitude and phase measurements in figures 16 and 17. (a) Rectangular nozzle with a straight exit, (b) single-bevelled nozzle, (c) double-bevelled nozzle.

the single- and double-bevelled nozzles exhibits a spanwise-oblique and spanwise-antisymmetric variation, respectively. The amplitude and phase information provided in this figure will be useful for assessing the potential damage from these nozzles as well as designing strategies including active control to minimize screech damage.

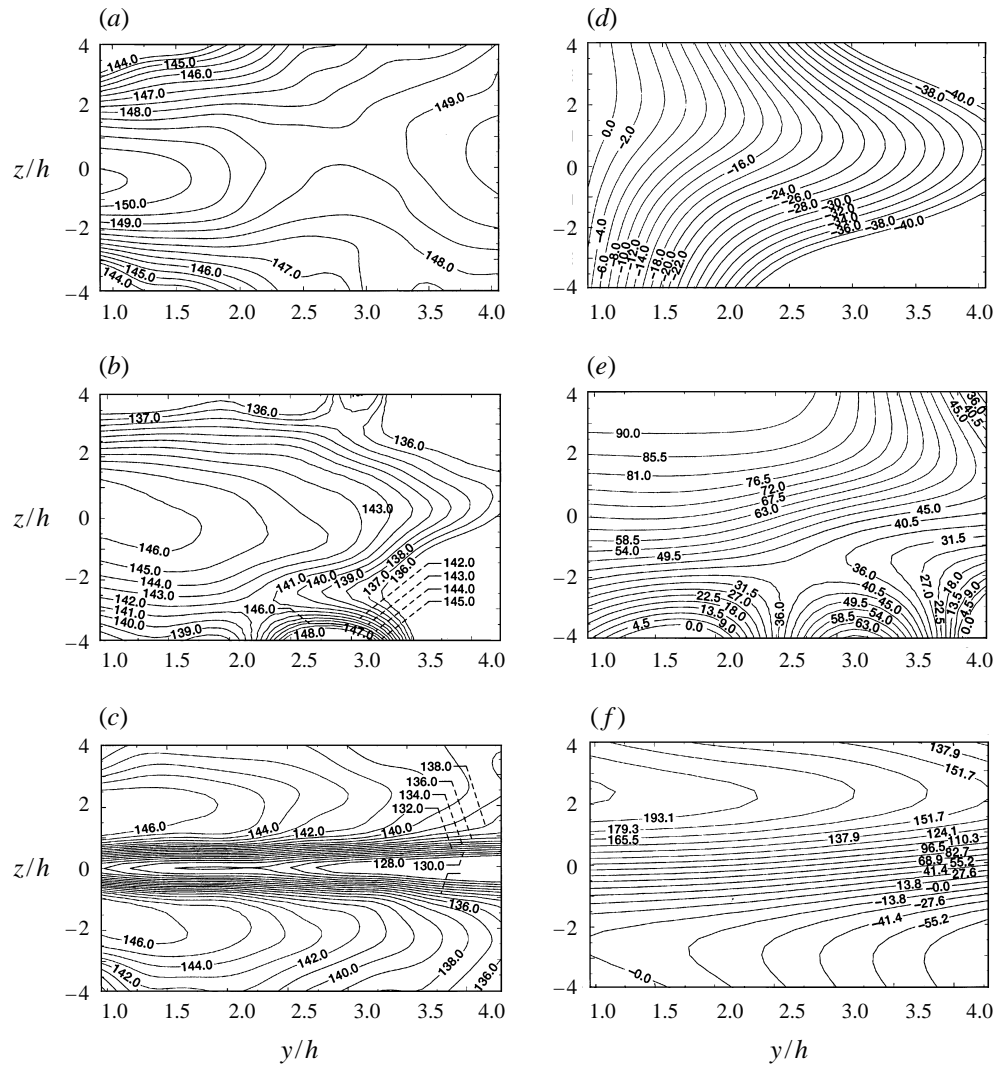


FIGURE 19. Screech sound pressure levels (dB) and isophase contours (in degrees) on the (y, z) -plane at $x/h = -1.25$. (a-c) Sound pressure levels, (d-f) isophase contours. (a, d) Jet from rectangular nozzle with a straight exit, (b, e) single-bevelled nozzle, (c, f) double-bevelled nozzle.

4. Concluding remarks

A detailed investigation was conducted on the screech tones from rectangular jets with spanwise-non-uniform exits. The primary motivation was the unexplored connection between shock-cell structure and screech modes. The secondary motivation was the lack of data on the screech characteristics of rectangular jets with spanwise-oblique shock-cell structures.

Three types of spanwise modes were observed: symmetric, antisymmetric, and oblique. The mode observed corresponded with the spanwise shock-cell structure. In cases where more than one mode was present, the technique presented here makes it possible to distinguish coexisting and mutually exclusive modes. The strengths of shocks 3 and 4 determined the screech source amplitude, and the unsteadiness of screech. Jets with spanwise-oblique shock-cell structures produced screech that was

generally weaker and more unsteady than jets with spanwise-uniform shock cells. In addition, jets from non-uniform nozzles screeched only over a limited Mach number range. The results presented provide answers to some of the issues relating to screech, and offer hope for a better understanding of screech and for tailoring shock-containing jets that minimize fatigue failure of aircraft components.

The work reported in this paper was an outgrowth of shear flow control efforts initiated by Dr E. J. Rice at the NASA Lewis Research Center. I would like to thank Professor C. K. W. Tam for many insightful discussions, and for his encouragement. Professor P. J. Morris clarified the refraction issue. Dr J. Panda's help with the schlieren photography is gratefully acknowledged.

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